

(b) *Blade passage frequency.*

(1) Each propeller blade produces noise and so the addition of blades will increase the noise. On the other hand, noise addition is logarithmic such that two equal noise levels added together will not result in doubling the noise emission but rather increase it only slightly. (For example, a noise level of 80 decibels (dB) combined with another noise level of 80 dB yields 83 dB and not 160 dB). As a consequence, a conversion to a slower turning three bladed propeller will have a greater effect in reducing noise by reducing helical tip Mach No. at the expense of a smaller increase in noise due to the increase in blade passage frequency.

(2) For rotary-wing aircraft, the influence of the main rotor as a noise source is usually much more dominant than the tail rotor. However, it is observed that during some manoeuvres, depending upon the location of the ground observer with respect to the rotorcraft, the tail rotor becomes dominant. This is recognizable as the tail rotor noise has a high frequency characteristic as compared to the relatively low frequency signature of the main rotor. Another phenomenon known as main rotor "blade slap", occurring during manoeuvres and during landing approaches, is particularly annoying. The above noise characteristics are common to all helicopters and for the designers of amateur built aircraft there may be little guidance available on means to reduce them other than to consider options available with respect to helical tip Mach No., shaft RPM, number of blades and blade shape. Trial and error methods will be necessary to optimize rotorcraft design with respect to performance and perhaps with some noise reduction.

(c) Blade shape and pitch. Blade shape and pitch are chosen for performance considerations and the noise reduction considerations must remain secondary. There are no general guidelines for noise reductions but rather the designer should experiment with various propeller designs. Constant speed propellers generally are quieter but cost and weight considerations remove them from the reach of most amateur aircraft builders.

(d) Engine RPM. A reduction in engine RPM will make a significant reduction in noise emissions. On the other hand, the engine noise often will be masked by the higher level noise emissions created by the propeller.

Generally speaking, a four stroke piston engine is quieter than an equivalent two stroke design, the latter of which gives a pronounced tonal characteristic.

(e) Exhaust noise. For piston engines, the introduction of efficient exhaust systems (mufflers) has greatly reduced this noise source to an acceptable minimum.

On the other hand for aircraft employing gas turbine engines, the exhaust noise is a dominant noise source. Designers/builders should consult with engine manufacturers about the feasibility of installing some form of acoustic treatment (hush kits) around the compressor and turbine cases and particularly around the exhaust nozzle.

(f) Transmission and Gearbox Noise. With the possible exception of rotorcraft, transmission noise is not normally significant to the ground observer. Transmissions and gearbox noise sources, while being considered for cabin noise reduction, are not considered dominant for the ground observer and can generally be ignored. If however, significant noise emissions are observed originating from either the transmission or gearbox and some form of acoustic liner is feasible to install, this should be considered.

(g) Airframe Noise. Airframe source noise is not recognized as a significant noise source for small aircraft and may be ignored.

5. Noise Control Considerations In Operation.

Pilots are reminded that while aircraft design is a significant factor, major contributors to effective noise control are the operational procedures.

Unnecessary low-flying, particularly at high power, near populated areas is not only dangerous, it is considered poor etiquette.

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